Amendments to the Claims:

This listing of claims will replace all prior versions, and listings, of claims in the application:

Listing of Claims:

Claims 1-5. (Cancelled)

6. (Currently Amended) A rotor for a gas turbine, having blades ~~arranged on the rotor and rotating together with the rotor, the blades~~ forming a blade ring and the blades within the blade ring being arranged at a different distance from one another and thus with a different blade pitch, a distance between the blades within the blade ring changing continuously or discontinuously in a circumferential direction and the distance between the blades within the blade ring dimensioned such that an imbalance is canceled out, wherein the rotor has ~~several~~ a first rotatable blade ~~rings~~ ring and a second rotatable blade ring arranged axially behind one another in the rotor, ~~and~~ wherein, within ~~each blade ring,~~ each of the first and second blade rings of the rotor the blades are arranged at a different distance from one another, and wherein a first arrangement of the different distances of the blades within the first blade ring of the rotor is different from a second arrangement of the different distances of the blades within the second blade ring of the rotor.

7. (Cancelled)

8. (Original) The rotor according to Claim 6, wherein the rotor is a turbine rotor or compressor rotor of a gas turbine.

9. (Original) The rotor according to Claim 8, wherein the gas turbine is a turbine of an airplane engine.

10. (Original) The rotor according to Claim 6, wherein the rotor is a fan rotor of a gas turbine.

11. (Original) The rotor according to Claim 10, wherein the gas turbine is a turbine of an airplane engine.

12. (Original) The rotor according to Claim 6, wherein the rotor is a blisk (bladed disk) or bling (bladed ring) of a gas turbine and wherein the blades are an integral component of the rotor.

13. (Original) The rotor according to Claim 12, wherein the gas turbine is a turbine of an airplane engine.

14. (Currently Amended) A rotor for a gas turbine, comprising:

a first rotatable blade ring in the rotor; and

a second rotatable blade ring in the rotor;

wherein the first and the second rotatable blade rings in the rotor are arranged axially one behind the other and wherein on each rotatable blade ring the blades are arranged such that a first blade is positioned at a first distance from a second blade and the second blade is positioned at a second distance from a third blade, wherein the first distance is different from the second distance;

and further wherein a first arrangement of the different distances of the blades within the first blade ring in the rotor is different from a second arrangement of the different distances of the blades within the second blade ring in the rotor.

15. (Original) The rotor according to Claim 14, wherein on each blade ring a fourth blade is positioned at the first distance from a fifth blade and the fifth blade is positioned at the second distance from a sixth blade and wherein the

first, second, and third blades are located diametrically opposed from the fourth, fifth, and sixth blades, respectively, on the blade ring.

16. (Cancelled)

17. (Currently Amended) A method for optimizing vibrations in a gas turbine engine, comprising the steps of:

rotating a first blade ring of a rotor;

rotating a second blade ring of the rotor, wherein the first and second rotating blade rings of the rotor are arranged axially behind one another;

wherein on each blade ring the blades are arranged such that a first blade is positioned at a first distance from a second blade and the second blade is positioned at a second distance from a third blade and wherein the first distance is different from the second distance and wherein a first arrangement of the different distances of the blades within the first blade ring in the rotor is different from a second arrangement of the different distances of the blades within the second blade ring in the rotor; and

continuously changing a frequency of a vibration of a stationary assembly of the gas turbine by rotating the first and second blade rings of the rotor.